

## The LMD Frost Point Hygrometer

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The “Laboratoire de Meteorologie Dynamique”, LMD, took part in various programs which imply in-situ measurement of the water vapor content in the middle and lower stratosphere and high troposphere.

Considering how difficult it is to make "in situ" measurement of the weak contents of water vapor in the stratosphere, a balloon-borne instrument was developed at LMD. **The first flight onboard Open Stratospheric Balloons (BSO)** took place in **1986** from the balloon launching station of CNES (Centre National d'Etudes Spatiales) in Aire sur l'Adour (Ovarlez J et al., 1987).

The aim was to develop an instrument to be carried by long duration balloons, the Infra-Red Montgolfiere (MIR), and **the first flights of the LMD hygrometer onboard MIR** took place in **1988** (Ovarlez J, 1991). Then, this hygrometer was modified in order to be carried **onboard aircraft in 1994** (Ovarlez J et al. 1996).

### 1. PROBLEM TO BE SOLVED

The vertical distribution of the atmospheric water vapor content, although very irregular in the troposphere, is characterized by the prevalence of a large decrease with altitude up to the tropopause level.

*In the stratosphere* the molar fraction of water vapor (usually indicated "volume mixing ratio") reaches few ppmv (part per million in volume), which correspond to frost-point temperatures ranging between  $-75^{\circ}\text{C}$  and  $-95^{\circ}\text{C}$ , for an atmospheric pressure ranging between 200 hPa. and 10 hPa., and an air temperature being between  $-40^{\circ}\text{C}$  and  $-90^{\circ}\text{C}$ . The weak water vapor contents, and the surrounding atmospheric conditions make in-situ measurements difficult. In the stratosphere, it is not sufficient to have an instrument able to measure these contents taking into account the environmental conditions, as it is necessary to take many precautions to avoid pollution from the balloon and from the payload.

Two design of balloons, implemented by the CNES, were used: Open Stratospheric Balloons (BSO), and Infra-red Montgolfieres (MIR). The BSO allows to carry out a vertical profile for a flight duration of a few hours and can carry heavy payloads, up to several hundred kilograms depending on the volume of the balloon. The MIR is a hot air balloon, like a montgolfiere, volume 36000 m<sup>3</sup>, and whose flight level depends on the collected radiation. During the night time, the balloon is heated by the terrestrial infra-red radiation; during day time, the additional heat provided by the sun induces the ascent of the balloon. The flight altitude of the MIR varies on average between 20 km during the day and 30 km during the night; the flight duration of these balloons can be more than one month. Their fall down is

due to their flight over significant cloudy area, during the night, which cut the balloon from its source of thermal energy, namely the ground. The payload is about 50 kg. Within the framework of the LMD campaigns devoted to water vapor measurements (program AMETHYSTE) the localization and the data collection were provided by the ARGOS system onboard satellite.

*In the troposphere*, it is necessary to have an instrument having a wide range of measurement as the water vapor mixing ratio vary between some ppmv and a few thousands of ppmv. In that part of the atmosphere, an aircraft was used: the German research aircraft, a Falcon, implemented by the DLR (Deutsches Zentrum für Luft und Raumfahrt), and whose altitude can reach 13 km.

## 2. INSTRUMENTAL CHOICES

### 2.1 The hygrometer using a thermoelectric cooler

To solve the various constraints occurring from the use of short and long duration balloons, a chilled mirror frost-point hygrometer using a thermoelectric Peltier device was built at LMD. The chilled mirror technique is simple and well known. The occurrence of dew or white frost on a cooled mirror is detected in an optical way. The temperature of the mirror is maintained at the temperature of condensation by a feedback loop controlling the cooling and the heating of the mirror. This temperature, the dew or frost point temperature is directly connected to the partial pressure of water vapor that characterizes the water vapor content of the atmosphere. The simultaneous measurement of the air pressure and air temperature allows the determination of the mixing ratio and of the relative humidity (Annex I).

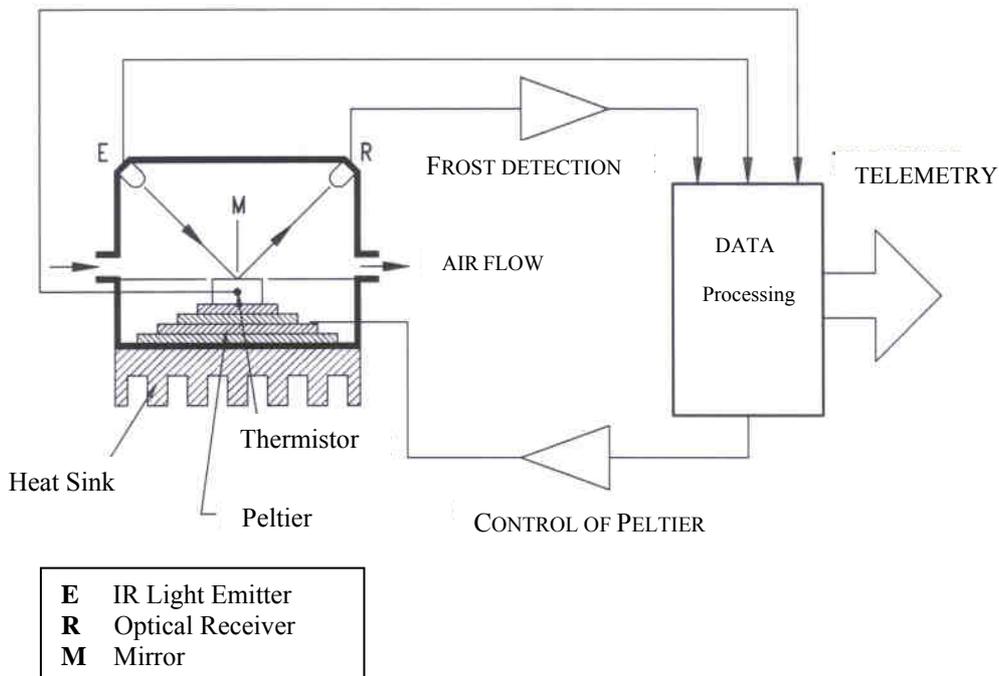


Figure 1 : Operating diagram of the frost point hygrometer

In the LMD instrument, the control of the temperature of the mirror is provided by a 4 stages thermoelectric module. The gilded copper mirror is directly stuck on the cold side of the thermoelectric module, and the temperature of the mirror is measured by a thermistor

embedded in the mirror, very near to its surface. A microprocessor makes the control of the drifts of the optical system and of the electronic components, their transfer and data processing towards the telemetry, as well as the regulation and control of the temperature of the mirror. The program, run by the microprocessor, and which controls the temperature of the mirror, uses more fuzzy logic than traditional proportional control loop. This is because, at the low frost-points to be measured, the way the white frost deposits on the mirror is not always reproducible, even under clearly defined conditions.

The thermistor embedded in the mirror is calibrated with respect to a platinum sensor associated to a multimeter both connected to the National Standards. In the same way, the air pressure sensor, essential to accurate mixing ratio determination, is calibrated with a transfer standard connected to the National Standards.

By using a Peltier thermoelectric module, we have an autonomous instrument which can be used on long duration balloon (MIR). However there are some limitations: the cooling efficiency of a thermoelectric module is strongly dependent on the temperature of the heat sink which is connected to the "hot side" of the module. As example, it is necessary to have  $-40^{\circ}\text{C}$  on the hot side to reach  $-95^{\circ}\text{C}$  on the mirror. One understands easily that measurements during the day are generally impossible, the instrument being strongly heated by solar radiation. On the other hand, during the night the heat sink is almost at the surrounding air temperature, which is generally between  $-40^{\circ}\text{C}$  and  $-80^{\circ}\text{C}$  according to the altitude of the balloon, and the cooling efficiency of the Peltier becomes sufficient to reach the low frost-points. However, for the MIR flights, it was necessary to develop, with the assistance of the CNES, a special heat sink looking upward to the sky, so that the instrument is cooled quickly when the sun lays down, i.e. when the MIR begins to go down towards its ceiling altitude of nighttime. The same kind of heat sink, a flat metal plate, was also used for measurements from BSO.

The instrument, with a Peltier module, flew onboard BSO until 1993 and onboard MIR until 1996. It was no more used after 1996, as a consequence of the end of the AMETHYSTE program (AMETHYSTE devoted to the water vapor measurements in the equatorial stratosphere by using the frost point hygrometer onboard MIR)

## **2.2 The hygrometer using cryogenic cooling**

From 1994, for the BSO and aircraft flights, the Peltier module was replaced by a cryogenic cooling system. This cooling system was selected so that the hygrometer could be used onboard aircraft, as the temperatures inside the aircraft are not low enough to have the necessary cooling efficiency of the Peltier module. In the new configuration of the hygrometer the mirror is stuck on a copper rode plunging in liquid nitrogen enclosed in a dewar container, especially adapted, which provides a permanent cooling. The control of the mirror temperature is made through the adjustment of the heating of the mirror which is provided by a resistive wire rolled up around the copper rode. The use of cryogenic cooling made it possible to decrease considerably the response time of the hygrometer, compared to the Peltier instrument. In addition, in the program run by the microprocessor, the algorithms for the control of the mirror temperature were set to take into account the different ranges for the frost point temperatures to be measured from aircraft on one hand, and from BSO on the other hand. Let us note that the optical system and the associated mechanical structure remain the same ones as for the Peltier instrument, as shown on the illustrations of annex IV. The mirror and the embedded thermistor are, them also, similar.

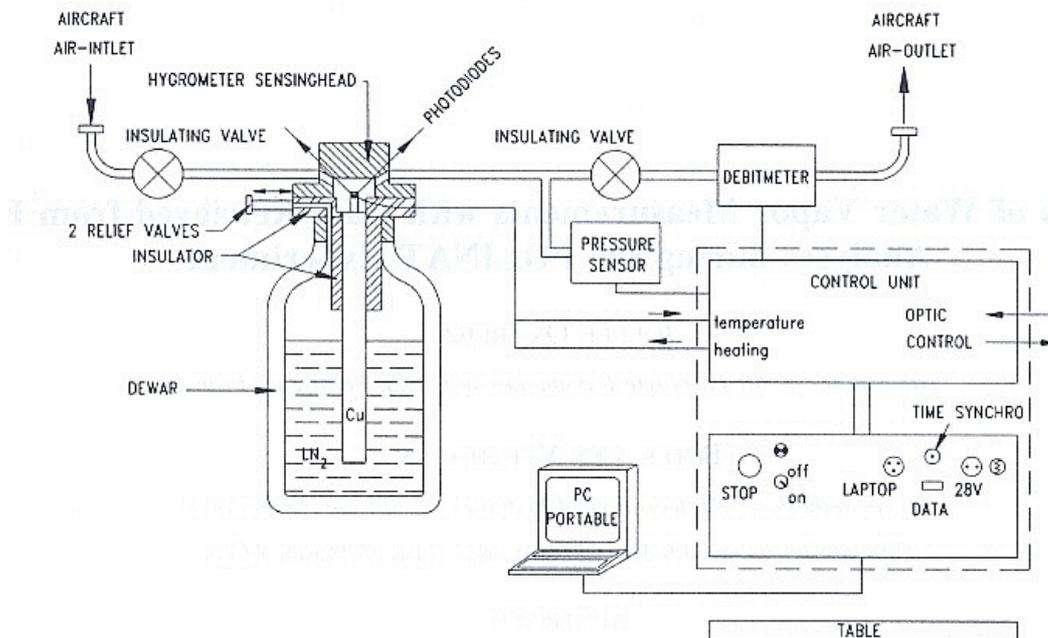


Figure 2: Diagram of the aircraft borne hygrometer

The development of the described instruments was made easier by the use of the LMD calibration system for hygrometers which allows the generation of environmental conditions, encountered in the stratosphere as well as in the middle and upper troposphere, for temperature, pressure, and moisture together (Ovarlez J, 1984, and annex I). This tool has been essential, particularly for the development of the program which controls the temperature of the mirror around the frost point through the microprocessor. In fact, this required hours and hours of simulations under different conditions. The hygrometers calibration system and the calibration facilities for temperature and pressure are the guarantee of reliable measurements.

### 3. SUMMARY

#### The instruments

The instruments developed at LMD are frost point hygrometers using chilled mirror technique and optical detection of the white frost on the mirror for:

- The long duration balloons (Infra Reds Montgolfieres, MIR)
- The Open Stratospheric Balloons (BSO)
- The research aircraft (since 1994)

2 systems for the cooling of the mirror are used:

- Cooling of the mirror by a thermoelectric module (Peltier) to have the autonomy necessary to the flights of long duration balloons (MIR).

This technique has been used for MIR flights until 1996, and also for BSO flights until 1993.

- Cryogenic Cooling for BSO and aircraft measurements from 1994

- These instruments allow direct measurement of the frost point temperature with an uncertainty  $2\sigma = 0.3\text{C}$  for the cryogenic hygrometer, and  $0.5\text{C}$  for the Peltier hygrometer.
- Joint measurement of the atmospheric pressure and temperature are made (Paroscientific sensor for air pressure; microbead thermistor VEECO Victory engineering Co. for the air temperature from balloons, and operational Rosemount/Goodrich temperature sensor for the measurements from aircraft).

One can thus calculate the hygrometric parameters (cf. annex I). The volume mixing ratio is then obtained with an uncertainty  $2\sigma = 3$  to  $6\%$  for the cryogenic hygrometer and  $10\%$  for the Peltier hygrometer

- The first balloon flight took place in 1986, a BSO flight starting from Aire sur l'Adour
- The first experiment from aircraft took place in 1994

### **The research programs:**

The LMD frost point hygrometer was used during many national and European research programs. The hygrometer payload, onboard balloons, is commonly known under the name of **ELHYSA** following an acronym used for the first research programs, as "Etude de L'Hygrométrie Stratosphérique". Then it was easy to keep this name as it is well known by our partners, particularly European partners.

Among the scientific programs the ELHYSA payload was associated to, there are:

- For MIR flights: AMETHYSTE (Application des Mir à l'ETude de l'HYgrométrie Stratosphérique) 1991, 1994.
- For BSO flights: CHEOPS (Chemistry of polar stratosphere) 1990; EASOE (European Arctic Stratospheric Ozone Experiment) 1991-92; TRAVERSE (Tracking vortex elements reaching southern Europe) 1993; SESAME (Second European Stratospheric arctic and mid latitude experiment) 1994-95; SABINE (Summer Arctic Balloon International Experiment) 1998; THESEO (Third European stratospheric experiment one ozone) 1999-2002; CIPA (Comprehensive investigation of polar stratospheric aerosols) 2000-02; Satellite validation from SAGE II (1989-90), ILAS (1997), ODIN (2001) and ENVISAT (2003-04).
- The aircraft hygrometer took part in POLINAT I and POLINAT 2 (Pollution from aircraft in the North Atlantic flight corridor) 1994-95, 1997 and INCA (Inter hemispheric difference in cirrus properties from anthropogenic emissions) 2000.

In the Annex III there is the list of the publications referring to the results of the measurements of the water vapor during various campaigns.

### **The data base:**

All the data from MIR and BSO experiments over the period 1988 - 2005, are public and are in the data base ETHER. To the water vapor measurements, are associated the air temperature and air pressure measurements, as well as the localization of the balloon (CNES data).

For the aircraft experiments, the data on ETHER include the measurements taken within the framework of POLINAT I and POLINAT 2.

The water vapor data from INCA are not in ETHER, because as measured in the occurrence of cirrus clouds, it requires the use of measurements of the other instruments, onboard the aircraft, to discriminate the presence or absence of cirrus. They are available in the base specific to INCA (<http://www.pa.op.dlr.de/inca/>)

ETHER:

<http://ether.ipsl.jussieu.fr/>

“Data/Services”

“select by experiment”

“Balloon-borne experiment " or “airborne experiment”

“ ELHYSA ”

"ELHYSA2" will comprise the data from the flights of the hygrometer obtained from October 2005

#### **4. THE PARTICIPANTS**

At LMD, many people contributed to the development of the instrument.

The main contributors are: Henri Ovarlez as Principle Technical Investigator, Joelle Ovarlez as Scientific Investigator, Jacques Capus, Jacques Crespin, Bertrand Gaubicher, and among the non permanent people, Eric Landais.

The mechanical and design department of LMD contributed, with more particularly Bernard Gillet and Olivier Bousquet.

The balloons were implemented by the CNES, and the aircraft by the DLR (Germany). We thank the operational balloon team at CNES, and particularly Pierre Malaterre for MIR flights, late Pierre Faucon for the success of all the balloon campaigns we took part with him, and Pierre Dedieu for meteorological support. We thank the operational team of DLR for the operation of the Falcon aircraft. We thank Ulrich Schumann and Hans Schlager (Institute für Physik für Atmosphäre, Oberpfafenhoffen, Germany) to have incited us to use the hygrometer on aircraft, as this enabled to improve the instrument considerably. The programs which allowed the development of the instrument and the participation in balloons and aircraft campaigns were financed by the BNM (Bureau National de Metrology), CNRS INSU, CNES, and the European Commission.

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***These are only the first publications on BSO, MIR, and aircraft measurements and on the calibration system. The whole bibliography is in annex II).***

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## **ANNEX I: DETERMINATION OF THE HYGROMETRIC PARAMETERS**

### **I GENERAL CASE**

I-1) The hygrometer measures the frost point temperature.

It is the temperature to which the air is saturated with respect to the ice .

The measurement of this temperature allows to calculate the water vapor pressure in the air,  $e'$  , which one identifies to the vapor pressure in pure phase  $e$  (in the field of interest the factor  $f$  such than  $e' = f \cdot e$  is 1 within 1/1000).

To determine the vapor  $e$  pressure, the OMM formulas (1984) are used - cf III -.

From  $e$  and the atmospheric pressure  $P$ , the volume mixing ratio is calculated [ I-2) ]

From  $e$  and the ambient temperature  $T_a$ , the relative humidity is calculated [ I-3) ]

I-2) The molar fraction of water vapor or "volume mixing ratio"  $x_v = n_v / (n_v + n_a)$  is the ratio of the number of mole of water vapor ( $n_v$ ) to the total number of moles of the humid air ( $n_v + n_a$ ).

One shows:  $x_v = e' / P = e / P$  ratio of the water vapor pressure,  $e$  , to the total pressure of the humid air,  $P$ .

It is without dimension. When expressed in  $10^{-6}$ , the unit is the ppmv; often indicated by  $r$

$e$  is inferred from the measurement of the frost point temperature by the hygrometer.  $P$  is measured by the air pressure sensor onboard the hygrometer payload.

I-3) the relative humidity

$$RH_i = 100 \frac{e}{e_i(T_a)} \quad \text{Relative Humidity with respect to ice, in \%}$$

$$RH_w = 100 \frac{e}{e_w(T_a)} \quad \text{Relative Humidity with respect to the liquid phase, in \%}$$

$e$  inferred from the measurement of the frost point temperature by the hygrometer.

$e_i(T_a)$  et  $e_w(T_a)$  saturation vapor pressure respectively above ice and liquid water at the air temperature  $T_a$  ( $e$ ,  $e_i$  et  $e_w$  calculated with the OMM or Sonntag formula –cf III)

### **II AIRCRAFT FLIGHTS**

**II-1) Frost point temperature and mixing ratio:**

The frost point temperature,  $T_{dh}$ , measured by the hygrometer is a little different from the frost point temperature,  $T_{de}$ , of the air crossed by the aircraft. This is because of the difference between the pressure  $P_h$  at the hygrometer sensing head and the pressure  $P_e$  of the air outside the aircraft, induced by the air inlet \* and the aircraft speed (cf figure 2).

On the other hand the volume mixing ratio,  $r$  , is maintained.

$$\text{that is to say } r = \frac{e_h}{P_h} = \frac{e_e}{P_e}$$

$e_e$  saturation vapour pressure at  $T_{de}$

$P_e$  static air pressure

$e_h$  saturation vapor pressure at  $T_{dh}$  (measured by the hygrometer)

$P_h$  pressure at the hygrometer head.

the volume mixing ratio  $r = \frac{e_h}{P_h}$  is inferred directly from measurements at the hygrometer.

\* a forward facing Rosemount-Goodrich Modified temperature housing

Determination of Tde:

$$e_e = e_h [ P_e/P_h ] = r \cdot P_e$$

OMM or Sonntag formula (cf III) cannot be used directly. Then we use an approximate formula to calculate Tde from  $e_e$  (Marti et Mausberger, 1993) :

$$T = \frac{A}{\text{Log}(10)e - B}$$

$$A = -2663.5 \pm 0.8$$

$$B = 12.537 \pm 0.011$$

$e$ , saturation vapor pressure above ice at temperature  $T$  ;  $e$  in pascals;  $T$  in K

In order to fit A and B, the hygrometer measurements from many aircraft flights have been used (POLINAT):

For RH<sub>i</sub> = 100,00% we should have Ta=Tde if the OMM formula (cf III), used to calculate RH<sub>i</sub>, and the Marti-Mausberger formula, used to calculate Tde, are strictly equivalent.

We keep the mean value for A, A= -2663.5, and fit B to have Ta= Tde, when the measurements give RH<sub>i</sub> = 100%

We found that B is a function of the frost point temperature :

$$B = 12.537 + [ 1.14 \cdot 10^{-4} (T) - 4,47 \cdot 10^{-4} ]$$

With  $T = T_{dh}$  (Here  $T$  is in C)

Between -30 et -60C, the corrective factor for B is between -0.004 et -0.007 (which is in the range of the corrective factor from Marti and Mausberger).

**II-2) The relative humidity:**

The air temperature, Ta, from the aircraft instrumentation is needed.

$$RH_i = 100 \frac{e_e}{e_i(Ta)} = 100 \frac{e_h}{e_i(Ta)} \cdot \frac{P_e}{P_h} = 100 r \frac{P_e}{e_i(Ta)} \quad \text{relative humidity/ice (\%)}$$

$$RH_w = 100 \frac{e_e}{e_w(Ta)} = 100 \frac{e_h}{e_w(Ta)} \cdot \frac{P_e}{P_h} = 100 r \frac{P_e}{e_w(Ta)} \quad \text{relative humidity/liquid phase (\%)}$$

$e_i$  (Ta) et  $e_w$  (Ta) saturation vapor pressure respectively above ice and above liquid water at temperature Ta

r volume mixing ratio inferred from the hygrometer measurements ( $r = \frac{e_h}{P_h}$ )

**III SATURATION VAPOR PRESSURE FORMULATION**

**ew with respect to liquid water, and ei with respect to ice, at temperature T.**

**OMM Formulas** (World Organization of Meteorology - WMO)

$$\begin{aligned} \text{Log} (10) e_w = & 10,79574 \cdot (1 - T_1/T) - 5,02800 \cdot \text{Log}(10)[T/T_1] \\ & + 1,50475 \cdot 10^{-4} [1 - 10^{-8,2969 (T/T_1 - 1)}] \\ & + 0,42873 \cdot 10^{-3} [10^{4,76955 (1 - T_1/T)} - 1] + 0,78614 \end{aligned}$$

$T_1 = 273,16$  ;  $e_w$  in hPa, et T in K

$$\text{Log} (10) e_i = -9,09685 \left( \frac{T_1}{T} - 1 \right) - 3,56654 \text{Log} (10) \left[ \frac{T_1}{T} \right] + 0,87682 \left[ 1 - \frac{T}{T_1} \right] + 0,78614$$

$T_1 = 273,16$  ;  $e_i$  in hPa, et T in K

**The European standards** (AFNOR 1994) recommend the Sonntag (1990) formula

$$\ln(e) = AT^{-1} + B + C T + D T^2 + E \ln(T)$$

For  $e_w$ , in the temperature range -100°C to +100°C:

A = -6096.9385, B = 21.2409642, C = -2.711193E-2, D = 1.673952E-5, E = 2.433502.

$e_w$  in Pa and T in K. [To have  $e_w$  in hPa, replace 21.2409642 by 16,635794]

For  $e_i$ , in the temperature range 0.01°C à -100°C :

A = -6024.5282, B = 29.32707, C = 1.0613868E-2, D = -1.3198825E-5, E = -0.49382577.

$e_i$  in Pa and T in K

[To have  $e_i$  in hPa replace 29.32707 by 24.7219]

### **Which formula to use?:**

In the case of balloon experiments, we always used the OMM formula.

Indeed, comparative calculations show

- A quasi equivalence of the 2 formula (OMM and Sonntag) for the saturation vapor pressures over ice. Thus there is no incidence, using one or the other formula, on the calculation of the mixing ratio which is the quantity generally used for the studies in stratosphere
- A light variation for the saturation vapor pressure over liquid water, corresponding to a difference of about  $\Delta RH = 0.5\%$  on the relative humidity  $RH_w$  for very humid air at low temperature. The field where the problem of the use of one or the other formula could arise is the cloud microphysics, the threshold for nucleation being highly dependant on the relative humidity. Thus, in the case of the aircraft flights for the study of the cirrus (INCA program), the relative humidity with respect to the liquid phase,  $RH_w$  is given, in our data base, from OMM formula and Sonntag one. It is in any case always necessary to specify the formula used (the more so as there are many other formulations)

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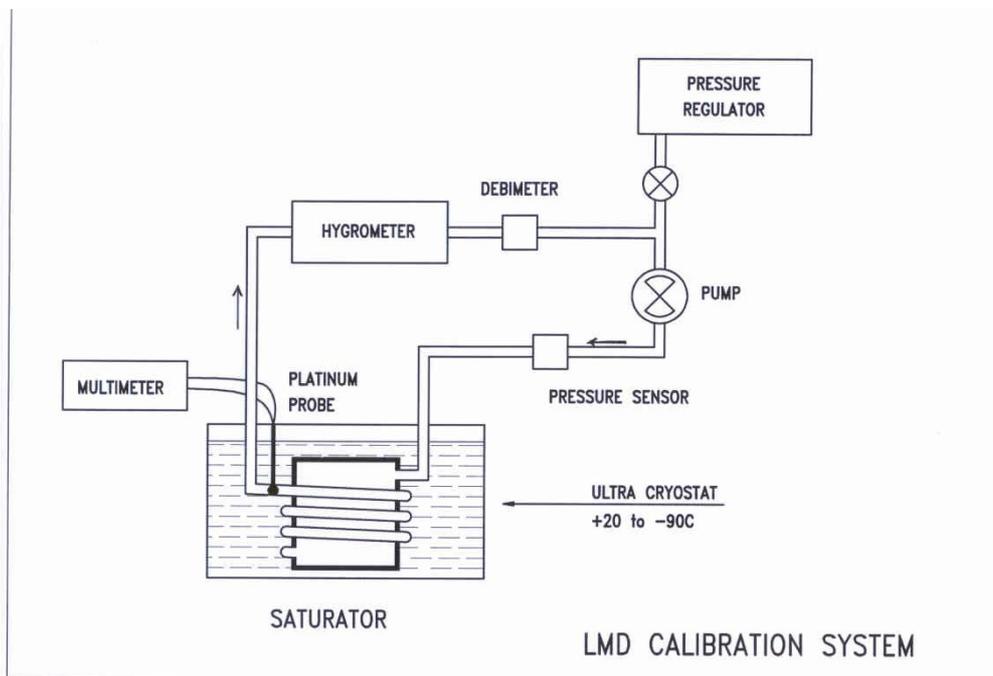
[The vapor pressure formulation for liquid water given in the 1988 edition has many typographic errors, so that the formula cannot be used! There is in the 2000 edition a "corrigendum to the 1988 edition" and .... it remains an error, hopefully with low consequences on the calculations). However the 1984 edition is correct!]

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## **ANNEX II**    **LMD CALIBRATION SYSTEM FOR LOW HUMIDITY**

In that device, as shown on the figure below, the air flows in a closed and tight loop through a saturator which, when the equilibrium is reached, fixes the dew point or frost point temperature of the air which penetrates into the hygrometer. The pressure of the "wet" air is fixed by a pressure regulator, and once the pressure is around the fixed value, the regulator is isolated and the pressure is controlled permanently through the pressure sensor. The saturator contains a small quantity of distilled water to allow the saturation of the air, and plunges in the methanol bath of an ultra-cryostat LAUDA which provides a very fine regulation of its temperature. Thus the temperature of the dewpoint or frost point of the humid air can be fixed between +10 and -90C in a pressure range between 20 and 1000 hPa. The temperature of the saturator is measured by a platinum sensor associated to a multimeter, both connected to the National References with the best levels of uncertainty.



The saturator and the connecting tubes in contact with the "wet air" are in polished stainless steel to reduce outgasing. ,

This unit is a simplified version of the system called "Two Temperatures System" initially built at LMD for the calibration of the sensors with respect to relative humidity, in a wide range of pressure and temperature.

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**ANNEX III:**

**PUBLICATIONS**

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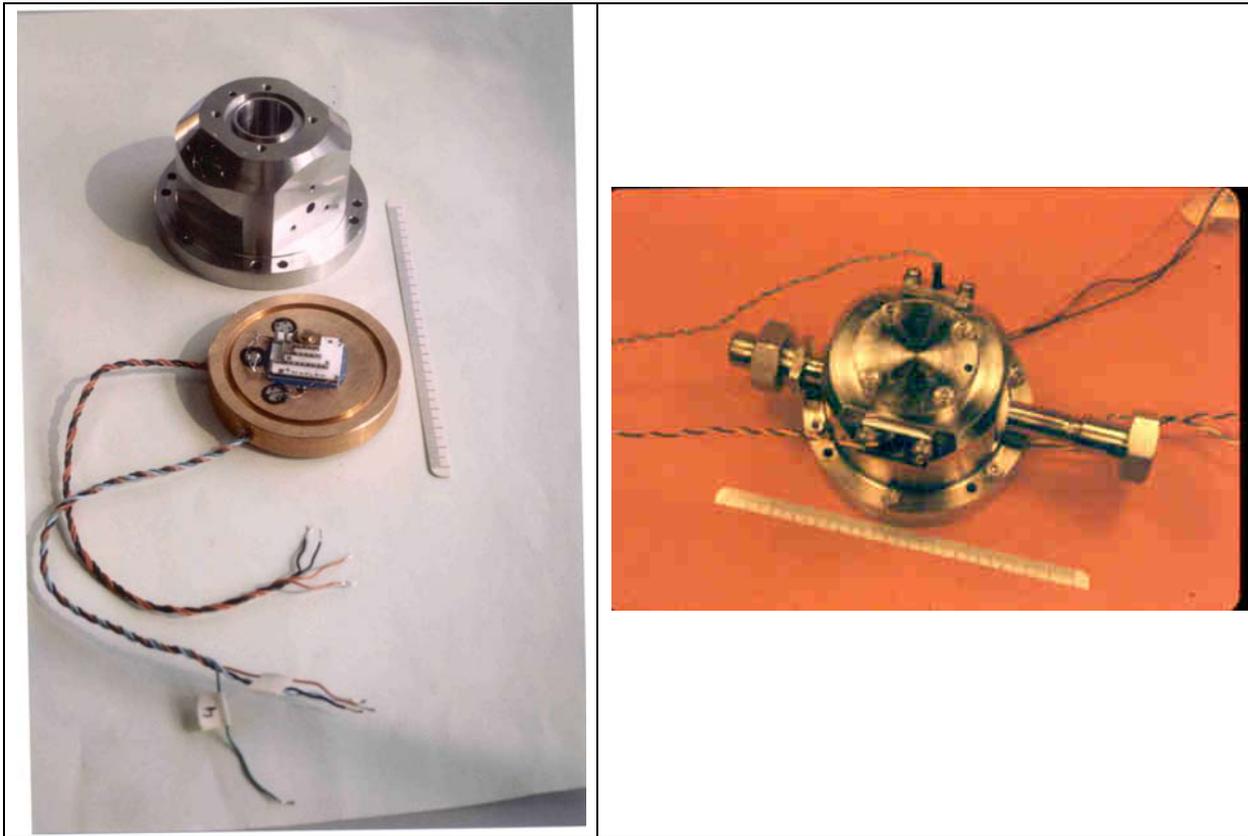
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**ANNEX IV**

**ILLUSTRATIONS**



**Hygrometer with Peltier :**

The mirror is stuck on a 4 stages Peltier module which is stuck on a copper heat sink. On the right, the picture shows the sensor head which is closed and tight. The holders of the optical transmitter and receiver are seen (cf figure 1).



**The hygrometer payload for MIR flights**

On the left side the holder for the air temperature sensor and for the air inlet tube  
On the right side, the ARGOS antenna.



**The hygrometer payload for MIR flights.**

The solar cells panels are seen.

The heat sink is above: it is an aluminium plate with an aluminised polystyrene support

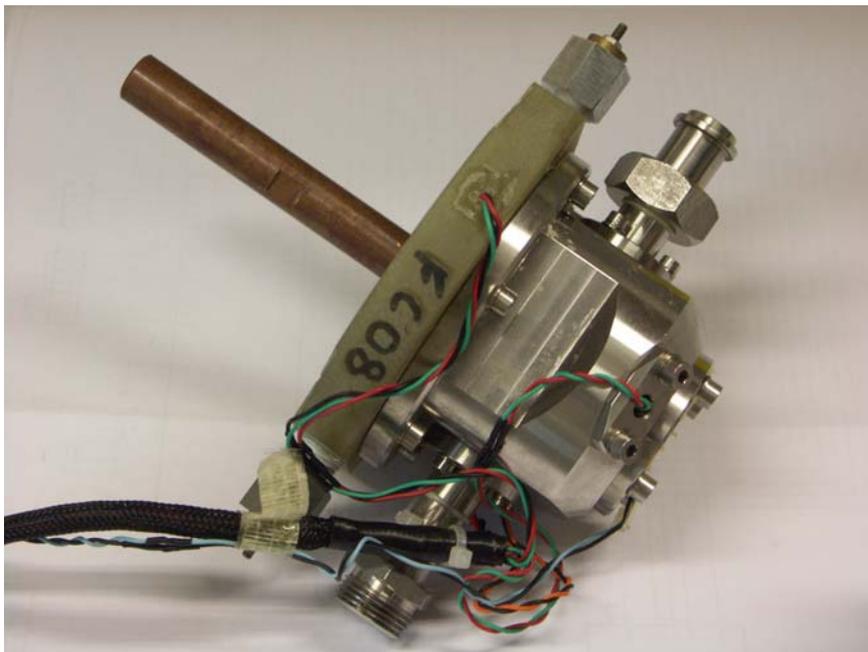
**The MIR** ready  
to be launched  
(Pretoria, South  
Africa)





**The cryogenic hygrometer :**

The mirror is on a thin rode. A resistance wire is wrapped around the rode



**The hygrometer head**

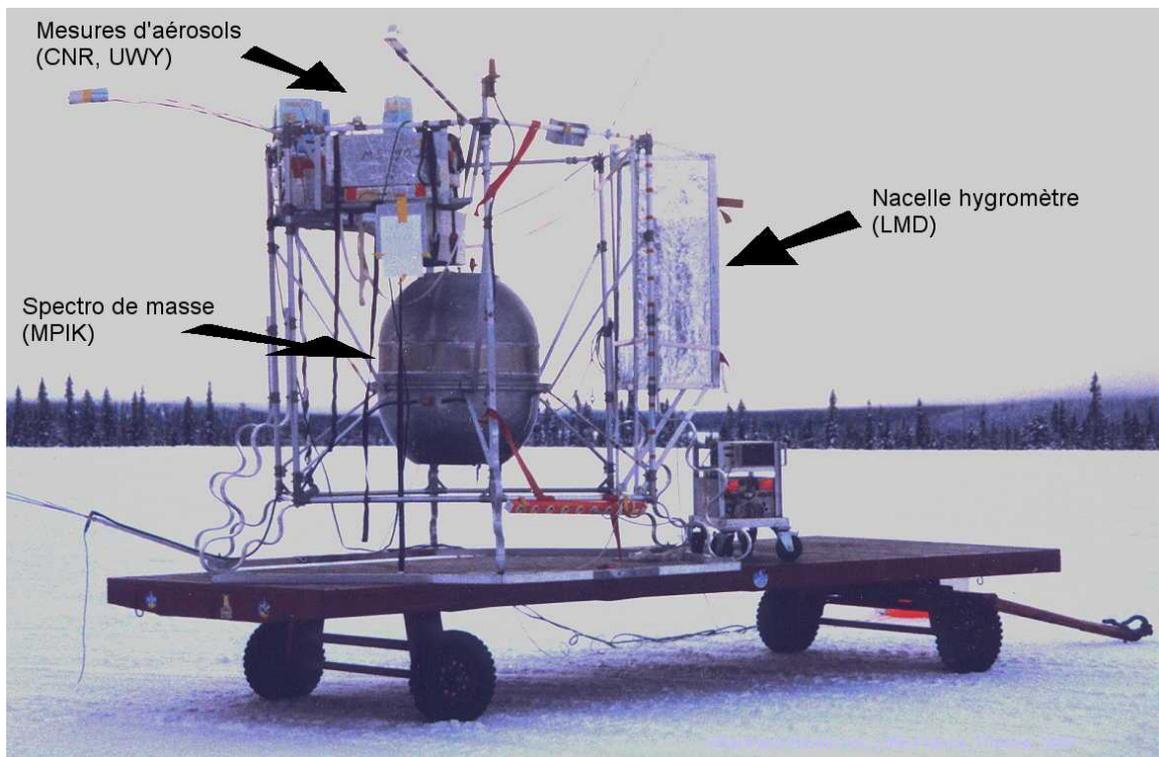
The tube to be plunged in liquid nitrogen is seen.

Except the tube, the hygrometer head is the same as the one of the Peltier hygrometer.

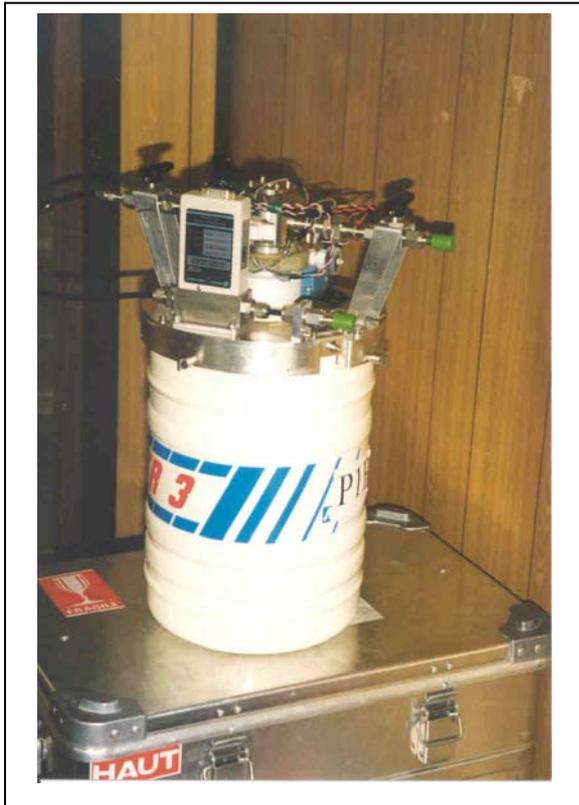
The mechanical connections seen are to link the hygrometer head to the calibration system with tight connections.



**The hygrometer payload for BSO, ELHYSA, here with Bertrand Gaubicher**



**The LMD hygrometer onboard the payload of Max Plank Institute, Heidelberg.**  
( PSC Analysis and CIPA european programs during THESEO)



**The hygrometer for aircraft**



**The cryogenic hygrometer inside the aircraft** with Jacques Capus  
(here the Falcon, research aircraft from DLR –Deutsches Zentrum für Luft und Raumfahrt)